

WHAT IS CLAIMED IS:

1. An annular platform for the nozzle of a low-pressure turbine in a turbomachine, said nozzle having a longitudinal axis and comprising at least one fixed vane disposed downstream from at least one moving blade of a high-pressure turbine, said platform comprising a downstream portion supporting said fixed vane radially defining an aerodynamic channel which extends longitudinally between a leading edge of said fixed vane and a trailing edge of said moving blade, said platform further comprising an upstream portion extending longitudinally beyond the leading edge of said fixed vane towards the trailing edge of said moving blade so as to lengthen said aerodynamic channel.
2. A platform according to claim 1, wherein said upstream portion includes a cooling circuit.
3. A platform according to claim 2, wherein said cooling circuit includes at least one cooling cavity extending longitudinally between an upstream end of said platform and the leading edge of the fixed vane.
4. A platform according to claim 3, wherein said cooling circuit further comprises air feed means for feeding said cavity, and air exhaust means for exhausting air from said cavity.
5. A platform according to claim 4, wherein said air exhaust means of said cavity comprise at least one hole opening out into said cavity and leading to the outside of said platform.
6. A platform according to claim 3, wherein said cavity is obtained by forming a recess in an outside portion of said platform and covering the recess with a closure device.

7. A platform according to claim 1, constituting a top platform of said high-pressure turbine nozzle.

5 8. A platform according to claim 4, constituting a top platform of said high-pressure turbine nozzle, and wherein said air feed means comprise at least one orifice opening out into an air manifold for cooling the fixed vane and leading into said cavity.

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9. A platform according to claim 8, wherein said air feed means comprise at least two orifices having different angles of inclination so as to distribute the cooling air uniformly within said cavity.

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10. A platform according to claim 1, constituting a bottom platform of said low-pressure turbine nozzle.

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11. A platform according to claim 4, constituting a bottom platform of said low-pressure turbine nozzle, and wherein said air feed means comprise an orifice passing through said platform for exhausting cooling air from said fixed vane.

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12. A platform according to claim 11, further comprising at least one deflector disposed parallel to said upstream portion of the platform so as to form an annular cooling channel between said deflector and said platform.

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13. A nozzle for a low-pressure turbine of a turbomachine, the nozzle comprising a plurality of fixed vanes supported by at least one top platform according to claim 7.

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14. A nozzle for a low-pressure turbine of a turbomachine, the nozzle comprising a plurality of fixed

vanes supported by at least one bottom platform according to claim 10.

15. A nozzle for a low-pressure turbine of a
5 turbomachine, the nozzle comprising a plurality of fixed vanes supported by at least one top platform according to claim 7 and by at least one bottom platform according to claim 10.